

MAS. 862 P. I. T.
Chapter 15 Problem Set

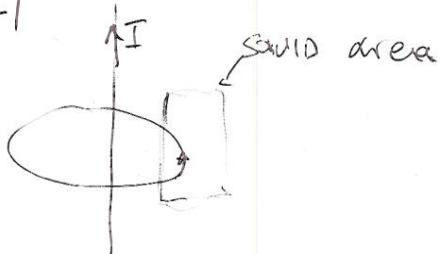
$$(15.1) E = 2E_F - 2E_C e^{-\frac{2}{N} V}$$

We cannot differentiate around $V=0$!

$$\text{Because } E(0) = 2E_F - 2E_C e^{\frac{2}{0}}$$

\rightarrow no perturbation theory.

(15.2)



$$H = \frac{I}{2nr} \Rightarrow \frac{B}{\mu_0} = \frac{I}{2nr}$$

$$\Rightarrow \frac{\phi}{A} = \frac{\mu_0 I}{2nr}$$

$$\Rightarrow r = \frac{\mu_0 I A}{2n \phi}$$

$$\text{In our case } r = \frac{4\pi \times 10^{-7} \text{ N} \cdot \text{A} \cdot 1 \text{ m}^2}{2 \cdot \pi \cdot 2.07 \times 10^{-7} \text{ Vs}^2} = 9.6 \times 10^3 \text{ m}$$

(15.3) Seconds in a month: 2592000 s , $R_{\text{Earth}} = 6.38 \times 10^6 \text{ m}$

So for Harrison's chronometer: $2592000 \text{ s} \cdot 2\pi \cdot 6.38 \times 10^6 \text{ m} \cdot \frac{10^{-5}}{84000 \text{ s}} \approx 11 \text{ km}$

while for G atom clock ($\frac{10^{-12}}{\text{1 day}}$) it is $\approx 1.2 \text{ mm}$

(15.5) (a) We want the centrifugal force to be equal to the gravity force:

$$\text{Diagram: A circle with a dot at the center. Two vectors, } F_c \text{ and } F_g, \text{ originate from the center. } F_c \text{ is tangent to the circle, } F_g \text{ is radial. } F_c = F_g \Rightarrow \frac{mv^2}{r} = \frac{GMm}{r^2} \Rightarrow v = \sqrt{\frac{GM}{r}}$$

$$v = \sqrt{\frac{6.67 \times 10^{-11} \text{ N} \cdot \text{m}^2 \cdot \text{kg}^{-2} \times 5.98 \times 10^{24} \text{ kg}}{2.02 \times 10^7 \text{ m} + 6.38 \times 10^6 \text{ m}}} \approx 3.8 \times 10^3 \text{ m/s}$$

orbit

earth radius

$$(15.8)(b) \quad N = \frac{2\pi r}{T} \Rightarrow T = \frac{2\pi r}{N} = \frac{2\pi r \times 2.66 \times 10^7 \text{ m}}{3.8 \times 10^3 \text{ m s}^{-1}} \approx 4.31 \times 10^4 \text{ s}$$

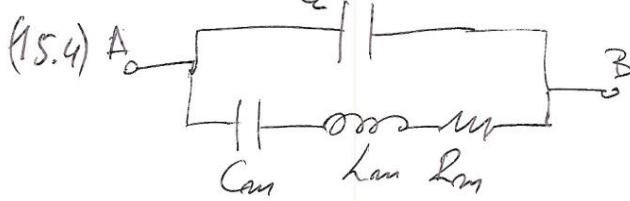
$$(c) \quad t_{\text{earth}} = 4.31 \times 10^4 \text{ s}$$

$$t_{\text{earth}} - t_{\text{sat}} = t_{\text{earth}} \cdot \left(1 - \sqrt{1 - \frac{v^2}{c^2}} \right) = 4.31 \times 10^4 \cdot \underbrace{\left(1 - \sqrt{1 - \frac{3.8 \times 10^3}{3 \times 10^8}} \right)}_{< 1 \text{ s} \Rightarrow \text{sat clock runs slower}} \approx 3.6 \times 10^{-6} \text{ s}$$

(d) likewise:

$$\begin{aligned} t_{\text{earth}} - t_{\text{sat}} &= t_{\text{earth}} \left(1 - \frac{1 - \frac{GM}{r^2 c^2}}{1 - \frac{GM}{rc^2}} \right) \\ &= 4.31 \times 10^4 \text{ s} \cdot \left(1 - \frac{1 - \frac{6.67 \times 10^{-11} \times 5.98 \times 10^{24}}{2.66 \times 10^7 \times (3 \times 10^8)^2}}{1 - \frac{6.67 \times 10^{-11} \times 5.98 \times 10^{24}}{6.38 \times 10^6 \times (3 \times 10^8)^2}} \right) \\ &\approx -22.9 \times 10^{-6} \text{ s} \end{aligned}$$

\Rightarrow sat clock runs faster



$$\text{We have } Z_{AB} = Z_c // Z_m = \frac{Z_c \cdot Z_m}{Z_c + Z_m} = \frac{\frac{1}{i\omega C_c} \cdot \left[\frac{1}{i\omega L_m} + i\omega L_m + R_m \right]}{\frac{1}{i\omega C_c} + \frac{1}{i\omega L_m} + i\omega L_m + R_m}$$

$$\begin{aligned} \text{In Laplace transform } Z(s) &= \frac{\frac{1}{sC_c} \left[\frac{1}{sL_m} + sL_m + R_m \right]}{\frac{1}{sC_c} + \frac{1}{sL_m} + sL_m + R_m} = \frac{sC_m \left[\frac{1}{sL_m} + s^2 C_m L_m + sC_m R_m \right]}{sC_m + sC_c + s^2 C_c C_m L_m + s^2 C_c C_m R_m} \\ &= \frac{s^2 C_m L_m + sC_m R_m + 1}{s^3 C_c C_m L_m + s^2 C_c C_m R_m + s(C_m + C_c)} \end{aligned}$$

Analysis & plots continue in attached matlab file and publication

(chap15prob4.m)